

FILE COPY
NO 4



N 62 50476

The Catholic University of America

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

REPORT No. 476

RELATION OF HYDROGEN AND METHANE TO CARBON MONOXIDE IN EXHAUST GASES FROM INTERNAL-COMBUSTION ENGINES

By HAROLD C. GERRISH and ARTHUR M. TESSMANN



THIS DOCUMENT ON LOAN FROM THE FILES OF

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS
LANGLEY AERONAUTICAL LABORATORY
LANGLEY FIELD, HAMPTON, VIRGINIA

RETURN TO THE ABOVE ADDRESS.

REQUESTS FOR PUBLICATIONS SHOULD BE ADDRESSED
AS FOLLOWS:

1933

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS
3512 H STREET, N. W.
WASHINGTON 25, D. C.

AERONAUTICAL SYMBOLS

1. FUNDAMENTAL AND DERIVED UNITS

	Symbol	Metric		English	
		Unit	Symbol	Unit	Symbol
Length-----	l	meter-----	m	foot (or mile)-----	ft. (or mi.)
Time-----	t	second-----	s	second (or hour)-----	sec. (or hr.)
Force-----	F	weight of 1 kilogram-----	kg	weight of 1 pound-----	lb.
Power-----	P	kg/m/s-----		horsepower-----	hp.
Speed-----		km/h-----	k.p.h.	mi./hr-----	m.p.h.
		m/s-----	m.p.s.	ft./sec-----	f.p.s.

2. GENERAL SYMBOLS, ETC.

W , Weight= mg	mk^2 , Moment of inertia (indicate axis of the
g , Standard acceleration of gravity= 9.80665	radius of gyration k , by proper sub-
m/s ² = 32.1740 ft./sec. ²	script).
m , Mass= $\frac{W}{g}$	S , Area.
ρ , Density (mass per unit volume).	S_w , Wing area, etc.
Standard density of dry air, 0.12497 (kg-m ⁻⁴	G , Gap.
s ²) at 15° C. and 750 mm= 0.002378	b , Span.
(lb.-ft. ⁻⁴ sec. ²).	c , Chord.
Specific weight of "standard" air, 1.2255	b^2
kg/m ³ = 0.07651 lb./ft. ³ .	\bar{S} , Aspect ratio.
	μ , Coefficient of viscosity.

3. AERODYNAMICAL SYMBOLS

V , True air speed.	Q , Resultant moment.
q , Dynamic (or impact) pressure= $\frac{1}{2}\rho V^2$.	Ω , Resultant angular velocity.
L , Lift, absolute coefficient $C_L=\frac{L}{qS}$	$\frac{Vl}{\mu}$, Reynolds Number, where l is a linear
D , Drag, absolute coefficient $C_D=\frac{D}{qS}$	dimension.
D_o , Profile drag, absolute coefficient $C_{D_o}=\frac{D_o}{qS}$	e.g., for a model airfoil 3 in. chord, 100
D_i , Induced drag, absolute coefficient $C_{D_i}=\frac{D_i}{qS}$	mi./hr. normal pressure, at 15° C., the
D_p , Parasite drag, absolute coefficient $C_{D_p}=\frac{D_p}{qS}$	corresponding number is 234,000;
C , Cross-wind force, absolute coefficient	or for a model of 10 cm chord 40 m/s,
$C_c=\frac{C}{qS}$	the corresponding number is 274,000.
R , Resultant force.	C_p , Center of pressure coefficient (ratio of
i_w , Angle of setting of wings (relative to	distance of c. p. from leading edge to
thrust line).	chord length).
i_s , Angle of stabilizer setting (relative to	α , Angle of attack.
thrust line).	ϵ , Angle of downwash.
	α_o , Angle of attack, infinite aspect ratio.
	α_i , Angle of attack, induced.
	α_a , Angle of attack, absolute.
	(Measured from zero lift position.)
	γ , Flight path angle.

REPORT No. 476

RELATION OF HYDROGEN AND METHANE TO CARBON MONOXIDE IN EXHAUST GASES FROM INTERNAL-COMBUSTION ENGINES

By HAROLD C. GERRISH and ARTHUR M. TESSMANN
Langley Memorial Aeronautical Laboratory

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

NAVY BUILDING, WASHINGTON, D.C.

(An independent Government establishment, created by act of Congress approved March 3, 1915, for the supervision and direction of the scientific study of the problems of flight. Its membership was increased to 15 by act approved March 2, 1929 (Public, No. 908, 70th Congress). It consists of members who are appointed by the President, all of whom serve as such without compensation.)

JOSEPH S. AMES, Ph.D., *Chairman*,
President, Johns Hopkins University, Baltimore, Md.
DAVID W. TAYLOR, D.Eng., *Vice Chairman*,
Washington, D.C.
CHARLES G. ABBOT, Sc.D.,
Secretary, Smithsonian Institution, Washington, D.C.
LYMAN J. BRIGGS, Ph.D.,
Director, Bureau of Standards, Washington, D.C.
ARTHUR B. COOK, Captain, United States Navy,
Assistant Chief, Bureau of Aeronautics, Navy Department, Washington, D.C.
WILLIAM F. DURAND, Ph.D.,
Professor Emeritus of Mechanical Engineering, Stanford University, California.
BENJAMIN D. FOULOIS, Major General, United States Army,
Chief of Air Corps, War Department, Washington, D.C.
HARRY F. GUGGENHEIM, M.A.,
Port Washington, Long Island, New York.
ERNEST J. KING, Rear Admiral, United States Navy,
Chief, Bureau of Aeronautics, Navy Department, Washington, D.C.
CHARLES A. LINDBERGH, LL.D.,
New York City.
WILLIAM P. MACCRACKEN, Jr., Ph.B.,
Washington, D.C.
CHARLES F. MARVIN, Sc.D.,
Chief, United States Weather Bureau, Washington, D.C.
HENRY C. PRATT, Brigadier General, United States Army,
Chief, Matériel Division, Air Corps, Wright Field, Dayton, Ohio.
EDWARD P. WARNER, M.S.,
Editor "Aviation," New York City.
ORVILLE WRIGHT, Sc.D.,
Dayton, Ohio.

GEORGE W. LEWIS, *Director of Aeronautical Research*.

JOHN F. VICTORY, *Secretary*.

HENRY J. E. REID, *Engineer in Charge, Langley Memorial Aeronautical Laboratory, Langley Field, Va.*

JOHN J. IDE, *Technical Assistant in Europe, Paris, France*.

EXECUTIVE COMMITTEE

JOSEPH S. AMES, *Chairman*.

DAVID W. TAYLOR, *Vice Chairman*.

CHARLES G. ABBOT.

LYMAN J. BRIGGS.

ARTHUR B. COOK.

BENJAMIN D. FOULOIS.

ERNEST J. KING.

CHARLES A. LINDBERGH.

WILLIAM P. MACCRACKEN, Jr.

CHARLES F. MARVIN.

HENRY C. PRATT.

EDWARD P. WARNER.

ORVILLE WRIGHT.

JOHN F. VICTORY, *Secretary*.

REPORT No. 476

RELATION OF HYDROGEN AND METHANE TO CARBON MONOXIDE IN EXHAUST GASES FROM INTERNAL-COMBUSTION ENGINES

By HAROLD C. GERRISH and ARTHUR M. TESSMANN

SUMMARY

The relation of hydrogen and methane to carbon monoxide in the exhaust gases from internal-combustion engines operating on standard-grade aviation gasoline, fighting-grade aviation gasoline, hydrogenated safety fuel, "Laboratory Diesel" fuel, and "Auto Diesel" fuel was determined by analysis of the exhaust gases. Two liquid-cooled single-cylinder spark-ignition, one 9-cylinder radial air-cooled spark-ignition, and two liquid-cooled single-cylinder compression-ignition engines were used.

The results of more than 100 exhaust-gas analyses showed that a linear relation existed between the carbon monoxide and the hydrogen found in the exhaust gas from engines using hydrocarbon fuels. A small amount of CH_4 was found to be always present in the exhaust gas, but the amount was independent of the air-fuel ratio and of the H-C ratio of the fuel. These relationships and the use of the Ostwald combustion diagram make available all the information of a complete exhaust-gas analysis when any two factors (CO_2 , CO, O_2 , or air-fuel ratio) are known. The preparation and use of an Ostwald combustion diagram are described. It is also shown that the air-fuel ratio supplied to the engine may be determined with a precision of ± 2 percent without measuring the air taken in by the engine, thus making the method of particular value for work outside the laboratory.

INTRODUCTION

The importance of the complete analysis of the exhaust gases from internal-combustion engines is not generally realized but, inasmuch as information relating to composition of the fuel used, air-fuel ratio, fuel wasted due to incomplete combustion, and in particular the carbon-monoxide content of the exhaust may be readily obtained from such analyses, it is evident that the analyses are of considerable importance, although little reliable data on exhaust-gas relationships are available. The value of a partial analysis of the exhaust gas in adjusting carburetors is clearly recognized today and several automatic instruments for analyzing exhaust gas have been made available to engine operators.

The simple type of Orsat apparatus has been largely used to determine the amounts of carbon dioxide (CO_2), carbon monoxide (CO), and oxygen (O_2) in the exhaust gas. The determination of the amount of CO is not as satisfactory as that of the other constituents because of the poor absorption characteristics of the solutions used. If relationships could be established between the O_2 and CO_2 of the exhaust gas and the other constituents (carbon monoxide (CO), hydrogen (H_2), and methane (CH_4)) the usefulness of the simple Orsat apparatus would be materially increased.

Little experimental evidence has been published on the correlation of CO, H_2 , and hydrocarbons in exhaust gases from internal-combustion engines. Fenning (reference 1), using gasoline in a single-cylinder sleeve-valve engine operating at 800 r.p.m., found that in the exhaust gases $\text{H}_2 = \frac{1}{4.62} \text{CO}^{1.38}$ and that CH_4 was almost entirely absent. He refers to an earlier experimenter, Ballentyne, who had found that in the exhaust gas of a gasoline engine $\text{H}_2 = 0.36 \text{CO}$ and $\text{CH}_4 = 0.12 \text{CO}$. Judge (reference 2, p. 103) states that the just-mentioned empirical relations are sufficiently accurate for estimating the percentages of H_2 and CH_4 present in the exhaust.

The purpose of this investigation was (1) to determine the relation between H_2 , CH_4 , and CO in the exhaust of 4-stroke-cycle engines using standard grade and fighting grade aviation gasoline, hydrogenated safety fuel, "Laboratory Diesel" fuel, and "Auto Diesel" fuel; and (2) to prepare combustion diagrams, the use of which would permit the estimation of CO, H_2 , and CH_4 in the exhaust gas from the determination of CO_2 and O_2 . This investigation was conducted by the National Advisory Committee for Aeronautics between 1931 and 1933.

APPARATUS AND METHOD

A modification of the Bureau of Mines gas-analysis apparatus was used during this investigation and is shown in figure 1. Special forms of pipettes developed by the Bureau of Standards (reference 3, p. 133) were used to obtain more efficient absorption.

The percentages of CO_2 and O_2 were determined in the usual manner by absorption in potassium hydroxide and alkaline pyrogallol, respectively. These solutions were prepared as recommended by Shepherd (reference 3, p. 145). Percentages of H_2 , CH_4 , and CO were computed from data obtained by the com-

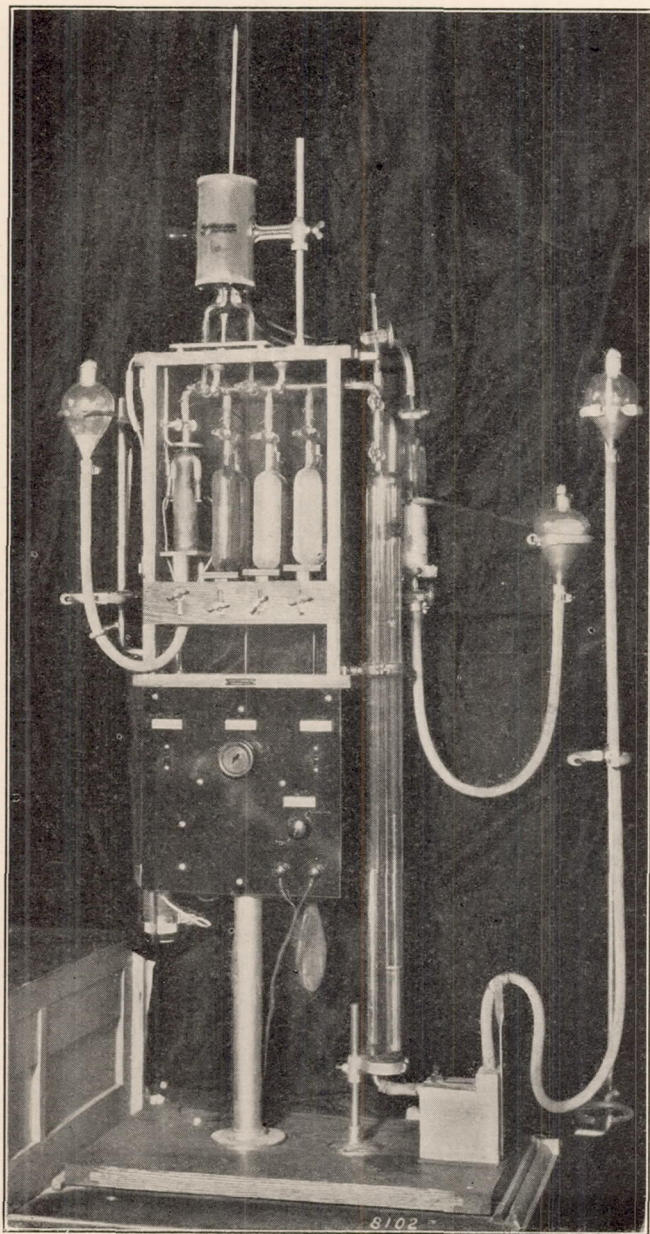


FIGURE 1.—Gas-analysis apparatus.

bustion of the residual gas and air in the slow-combustion pipette (reference 3, p. 162). Tests with fuming sulphuric acid showed no unsaturated hydrocarbons present.

Gas samples were obtained by inserting a steel tube into the exhaust pipe immediately behind the exhaust valve of the single-cylinder test engines and approximately 5 feet from the open end of the exhaust pipe of the 9-cylinder radial engine. Several minutes after the engine had attained a stable condition, the samples

were collected in glass sampling tubes. Samples were first taken by displacement of mercury, but later evacuated sampling tubes were used, employing only those that would hold an absolute pressure of 0.1 mm of mercury for several hours. Samples by either method gave consistent results. Portions of several check samples were analyzed immediately and at intervals of 1 and 2 weeks and no difference was found in the analyses.

In this investigation four single-cylinder 4-stroke liquid-cooled test engines and one 9-cylinder radial air-cooled engine were used. Table I shows the engine test conditions and the fuels used. The distillation curves (A.S.T.M.) for the fuels are shown in figure 2.

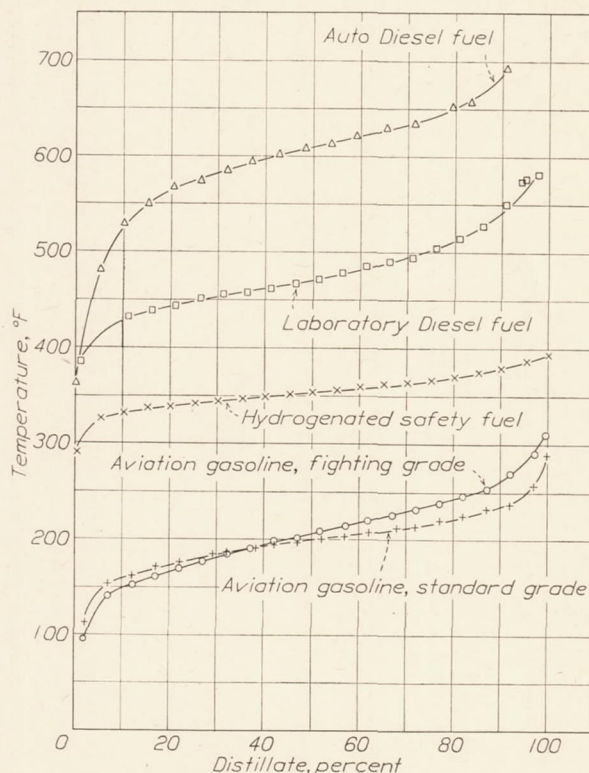


FIGURE 2.—Distillation curves (A.S.T.M.).

The H-C ratio of hydrogenated safety fuel was given by the manufacturer as 0.130. As the gasolines conformed to the Air Service specifications, they have been considered to have an H-C ratio of approximately 0.175 (reference 4, p. 461).

Fuel consumption was measured directly by weight, and air consumption by means of an 80-cubic-foot-capacity gasometer. The start and stop of the gasometer were synchronized electrically with the engine revolution counter and a stop watch.

RESULTS AND DISCUSSION

Relation of H_2 and CH_4 to CO .—The percentages of H_2 , CH_4 , and CO found in the exhaust gases of the engines using standard grade and fighting grade aviation gasolines, Auto Diesel fuel, and Laboratory Diesel fuel are plotted in figure 3. A mean curve drawn

through all the data gives a reasonably accurate approximation for the H-CO ratio. The test points for the various engines and fuels lie surprisingly near a mean curve for all the data. However, there is a small deviation of the mean curve for each set of data from the mean curve for all the data, indicating a slight shift in the H-CO ratio with changes in fuels and engines. The maximum deviation of the experimental points from this mean curve is about 20 percent for the large quantities of CO. The relation of H_2 to CO, considering all the test points, is $H_2 = 0.51CO$, whereas CH_4 was found to be constant, $CH_4 = 0.22$ percent.

In figure 3 a comparison is also made between the results obtained when using a carburetor and when using a fuel-injection system with spark ignition. It is evident that the relation of H_2 to CO is unaffected by the manner in which the fuel is introduced into the engine cylinder.

The experimental data establishing the relationship between H_2 and CO in the exhaust gas from an engine using hydrogenated safety fuel are given in figure 4. The relationship was found to be $H_2 = 0.33CO$. The

line, Auto Diesel fuel, and Laboratory Diesel fuel. It will be seen that a linear relation exists between these gases for lean mixtures, whereas for rich mixtures there is practically a constant amount of O_2

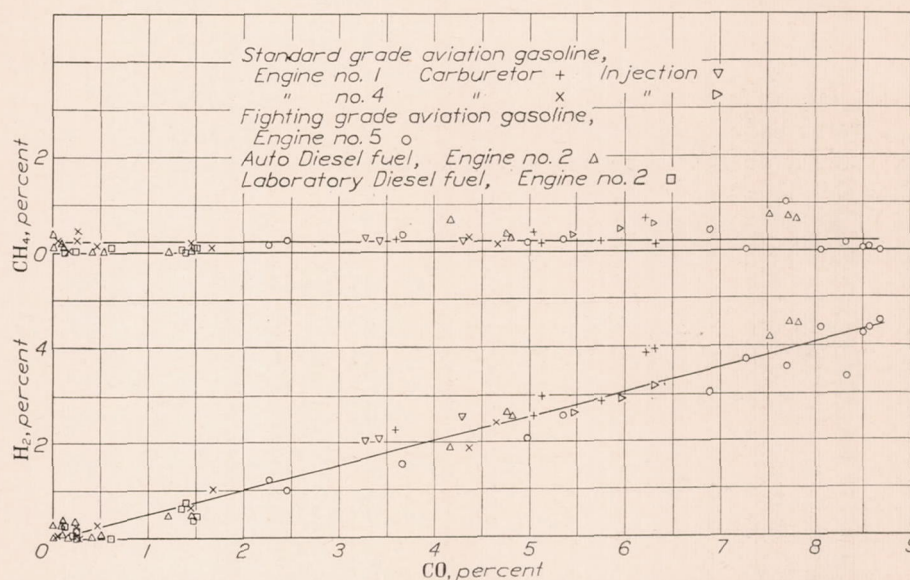


FIGURE 3.—Relation of CO, H₂, and CH₄ in exhaust gases of engines using standard and fighting grades of aviation gasolines, Auto Diesel fuel, and Laboratory Diesel fuel. The amount of CH₄ is constant, 0.22 percent. The amount of H₂ varies linearly with CO, H₂=0.51 CO.

present. It will be noted that for lean mixtures the experimental and theoretical results agree. For rich mixtures Dicksee (reference 5) found no free oxygen present. Fenning found oxygen present when using rich mixtures in a single-cylinder sleeve-valve test engine, but found none in his bomb experiments. He states (reference 1, p. 204) that the free oxygen found for rich mixtures was probably due to leakage past the sleeve valves during compression. Best (reference 6) analyzed the exhaust gases from separate exhaust stacks of a 6-cylinder engine operating at 1,300 r.p.m., full throttle, and with an air-fuel ratio of 11.9 (by measurement of air and fuel). He found oxygen present in each stack. It is believed that the presence of free oxygen with rich mixtures may be attributed either to the lack of a perfectly

homogeneous charge in the engine cylinder or to the dissociation of the oxides of nitrogen that are formed during the combustion process.

It is evident that if the quantity of O_2 present in the exhaust for lean mixtures is known the corresponding quantity of CO_2 can be readily determined from the curve shown in figure 5. However, for rich or near-rich mixtures it would be necessary to measure both gases, for in this range the O_2 content is practically constant.

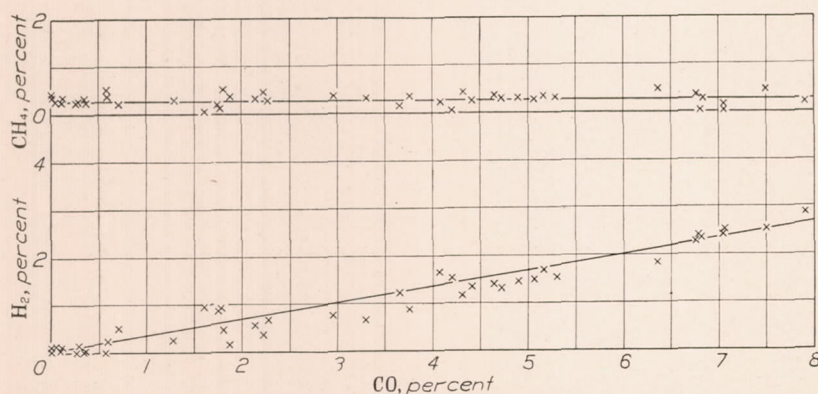


FIGURE 4.—Relation of CO, H₂, and CH₄ in exhaust gases of engines using hydrogenated safety fuel. The amount of CH₄ is constant, 0.27 percent. The amount of H₂ varies linearly with CO, H₂=0.33CO.

amount of CH_4 was found to be constant and equal to 0.27 percent.

No satisfactory explanation for the constancy of the percentage of CH_4 in the exhaust gas of spark-ignition and compression-ignition engines, irrespective of the load or type of fuel employed, is available at this time.

Figure 5 shows the relation of O_2 and CO_2 in the exhausts of three single-cylinder test engines and one 9-cylinder radial engine. The fuels used were standard-grade aviation gasoline, fighting grade aviation gaso-

The Ostwald combustion diagram shows in graphical form the theoretical relationships among the products of combustion of hydrocarbons. The interdependence of CO_2 , O_2 , CO , and air-fuel ratio is given and it is possible to determine CO and air-fuel ratio when the values for CO_2 and O_2 are known. Such diagrams are shown in figures 6 and 7 and their development is given in the appendix. The simplest type of Orsat apparatus enabling determinations of CO_2 and O_2 to be made is all that is needed in conjunction with an Ostwald diagram and knowledge of the relation of H_2 and CH_4 to CO to give a complete analysis of exhaust gas with an accuracy comparable to that of direct

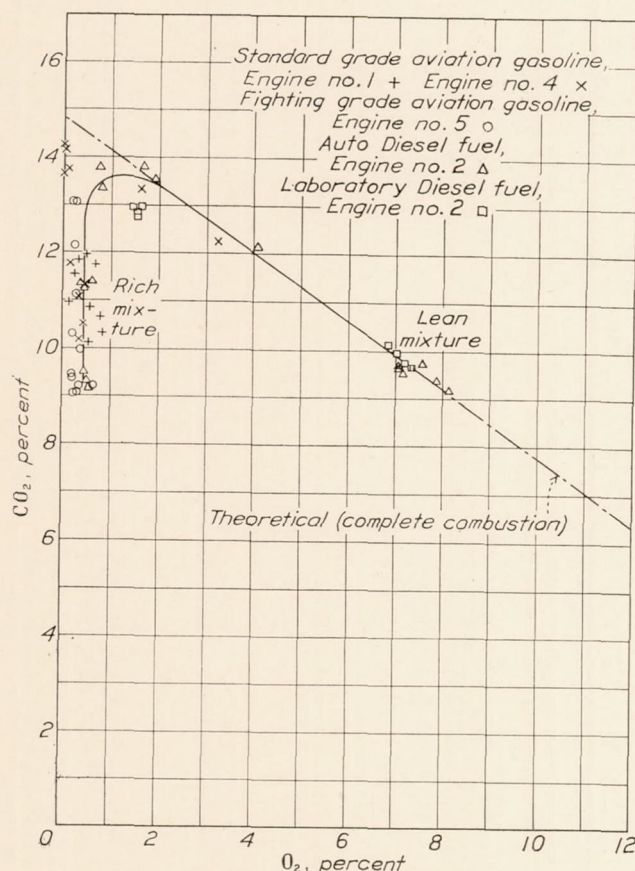


FIGURE 5.—Relation between CO_2 and O_2 in engine exhaust gases.

measurement. This fact should cause the diagrams to be of considerable value to engineers interested in the composition of exhaust gas from internal-combustion engines.

Figure 6 is an Ostwald combustion diagram for standard and fighting grade aviation gasolines having H-C ratios of 0.175. It is also applicable to Auto Diesel and Laboratory Diesel fuels since the H-C ratio of these fuels was found to be approximately the same as for the gasolines. An Ostwald combustion diagram for hydrogenated safety fuel (H-C ratio of 0.128) is shown in figure 7. The differences in the relationships of the two charts are due to the H-C ratio of safety fuel being different from that of the other fuels.

The dotted lines in figure 6 illustrate the use of the diagram. It is readily apparent that for an exhaust gas containing 10.5 percent CO_2 and 2.0 percent O_2 the CO content will be 4.5 percent and η will be 1.09, η being the reciprocal of the excess-air coefficient. The percentages of H_2 can be determined from the relation $\text{H}_2 = 0.51\text{CO}$. The amount of CH_4 is constant and equal to 0.22 percent.

The agreement of the values for air-fuel ratio obtained from the Ostwald diagram and those derived by calculation from the results of complete exhaust-gas analyses is shown in figure 8. The agreement of the values for CO obtained by the two methods is good and is shown in figure 9.

In figure 10 air-fuel ratios determined from analyses of the exhaust gases are compared to air-fuel ratios supplied to the engine and determined by weighing the fuel and measuring the air. For rich mixtures the air-fuel ratios found from exhaust-gas analyses were, for all engines tested, somewhat greater than those supplied to the engine, which is probably due to incomplete combustion of the fuel in the cylinder with consequent carbon formation not accounted for in the exhaust-gas analysis. The determination of the percentages of CO_2 and O_2 in the exhaust gas and the use of the Ostwald combustion diagram to obtain the exhaust air-fuel ratio make possible, by means of figure 10, the determination of the air-fuel ratio supplied to the engine with a precision of ± 2 percent without measuring the air taken in by the engine.

In table II is shown the composition of the fuels used, as supplied by the manufacturers or reported in the literature and as determined from exhaust-gas analyses. The averages of laboratory analyses are chiefly those of rich mixtures. It seems that values computed from mixtures a little leaner than the theoretically correct air-fuel ratios would have been more accurate than any others since deposition of carbon is more likely to occur in very lean and in rich mixtures.

The agreement between reported values and values computed from laboratory gas analyses is quite good for the gasolines and the safety fuel. The results from gas analyses of Laboratory Diesel fuel indicate it to have approximately the same composition as Auto Diesel fuel.

Table II and the relations found between H and CO for the different fuels indicate that there is a definite connection between the H-C ratio of the fuel and the H-CO ratio of the exhaust gases of internal-combustion engines because an increase in the H-C ratio of the fuel results in an increase in the H-CO ratio of the exhaust gas.

CONCLUSIONS

The experimental data presented in this report indicate that:

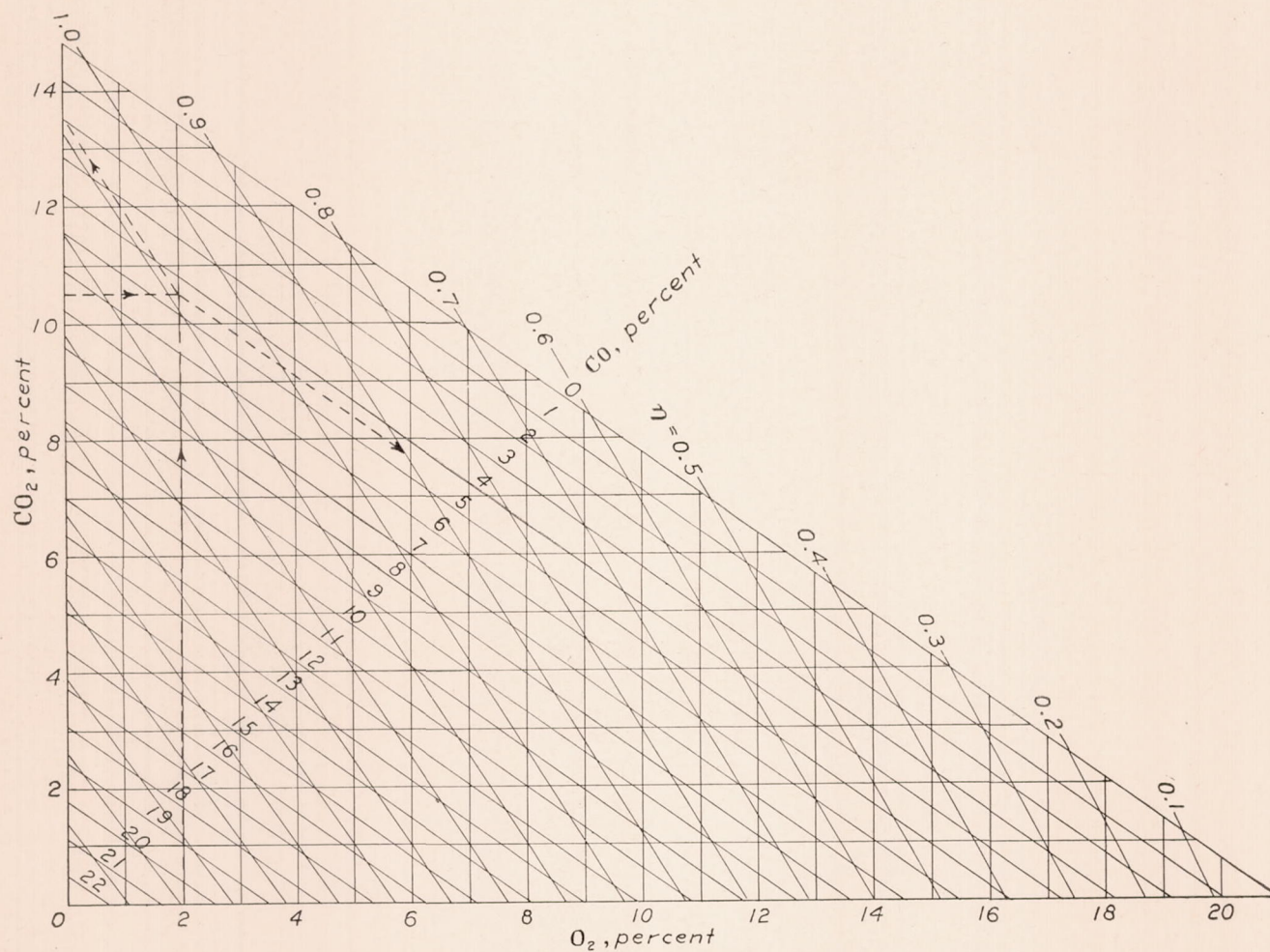


FIGURE 6.—Ostwald combustion diagram for standard and fighting grades of aviation gasolines, Auto Diesel fuel, and Laboratory Diesel fuel. The amount of CH_4 is constant, 0.22 percent. The amount of H_2 varies linearly with CO, $\text{H}_2=0.51\text{CO}$.

NOTE.—Excess-air coefficient $=1/\eta$; percent excess air $=100(1/\eta-1)$; air-fuel ratio $=14.89/\eta$.

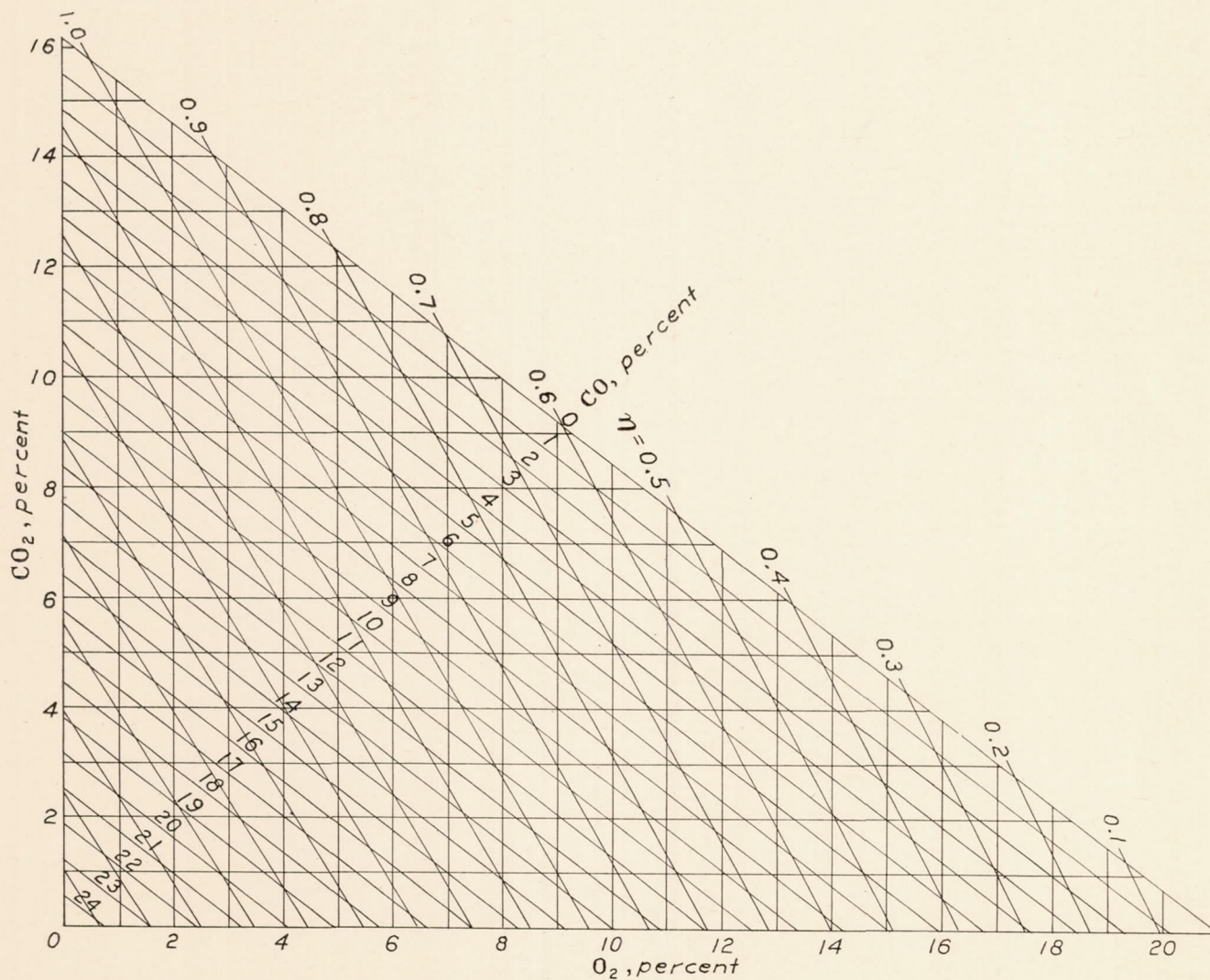


FIGURE 7.—Ostwald combustion diagram for hydrogenated safety fuel. The amount of CH_4 is constant, 0.27 percent. The amount of H_2 varies linearly with CO , $\text{H}_2 = 0.33\text{CO}$.

NOTE.—Excess-air coefficient $= 1/\eta$; percent excess air $= 100(1/\eta - 1)$; air-fuel ratio $= 14.09/\eta$.

1. The exhaust gases from internal-combustion engines burning hydrocarbon fuels contain H_2 and CO

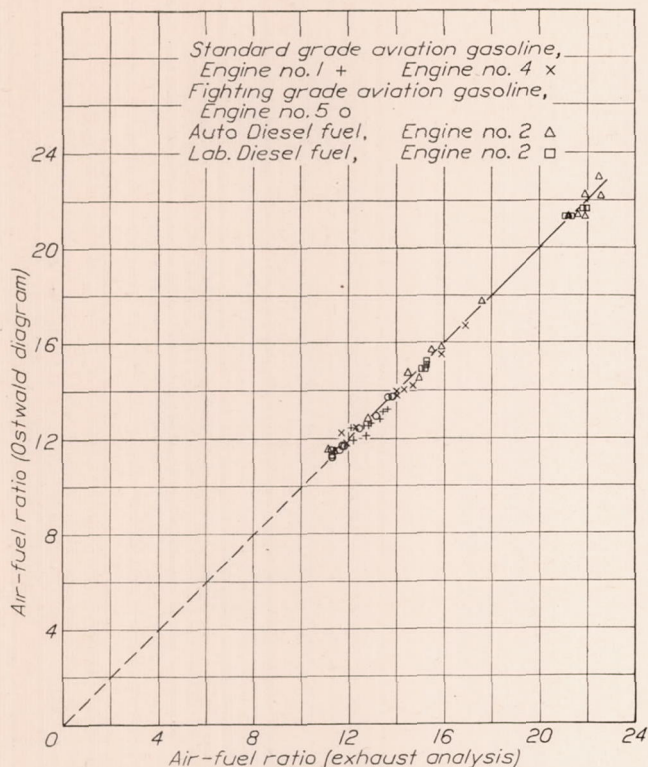


FIGURE 8.—Agreement of Ostwald combustion diagram and experimental values of air-fuel ratio.

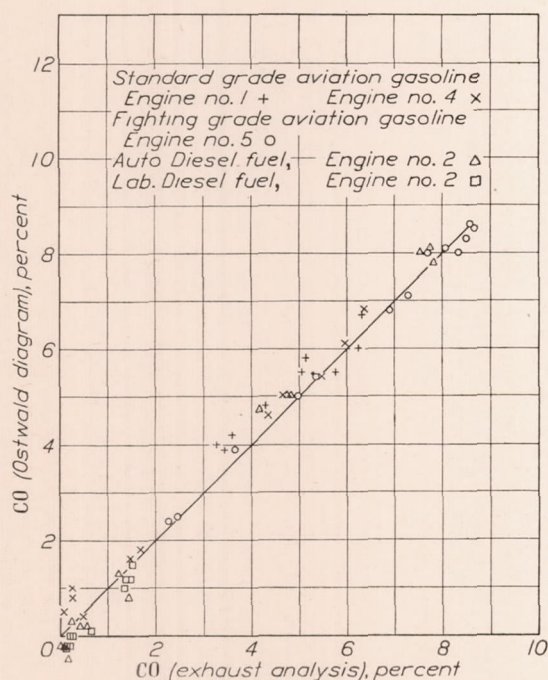


FIGURE 9.—Agreement of Ostwald combustion diagram and experimental values of CO.

in definite proportions, the ratio depending on the ratio of hydrogen to carbon present in the fuel. The

relation found for standard grade and fighting grade aviation gasolines, Auto Diesel fuel, and Laboratory Diesel fuel was $H_2 = 0.51CO$; for hydrogenated safety fuel the relation was $H_2 = 0.33CO$.

2. A small amount of CH_4 was found to be always present in the exhaust gas, but the amount was independent of the air-fuel ratio and of the H-C ratio of the fuel.

3. Oxygen was also found in the exhaust gas from internal-combustion engines using hydrocarbon fuels even when the fuel was in excess of that required for complete combustion.

4. The determination of any two components (CO_2 , CO, O_2 , air-fuel ratio), employment of the H-CO

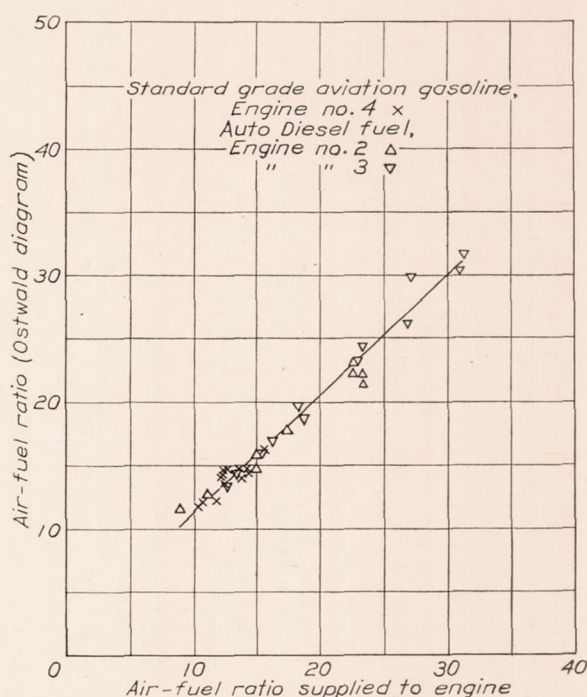


FIGURE 10.—Relation of air-fuel ratio supplied and air-fuel ratio in the exhaust gases of spark-ignition and compression-ignition engines.

relationship, and the use of the Ostwald combustion diagram provide a rapid means of obtaining the quantities of the products of combustion in the exhaust gas of internal-combustion engines.

5. The air-fuel ratio supplied to the engine may be determined from the exhaust-gas analysis with a precision of ± 2 percent without measuring the air taken in by the engine, thus rendering the method particularly valuable for work outside the laboratory.

LANGLEY MEMORIAL AERONAUTICAL LABORATORY,
NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS,
LANGLEY FIELD, VA., August 17, 1933.

APPENDIX

DEVELOPMENT OF AN OSTWALD COMBUSTION DIAGRAM

The following development of the Ostwald diagram for the combustion of a hydrocarbon, when the products of combustion contain the gases H_2 , H_2O , O_2 , N_2 , CH_4 , CO_2 , and CO , is similar to that of Theodorsen (reference 7).

The relative weights of carbon and hydrogen appearing in the exhaust as products of combustion are

$$C = 12(CO_2 + CO + CH_4)$$

$$H = 2.015(H_2 + H_2O + 2CH_4)$$

$$\text{Then } \frac{H}{C} = \frac{2.015(H_2 + H_2O + 2CH_4)}{12(CO_2 + CO + CH_4)} = K \quad (1)$$

where K is the weight ratio of the hydrogen to carbon and the chemical symbols refer to percentage by volume.

From equation (1):

$$H_2O = 5.955K(CO_2 + CO + CH_4) - H_2 - 2CH_4 \quad (2)$$

Considering dry air to contain 20.9 percent O_2 and 79.1 percent N_2 by volume, then

$$N_2 = \frac{79.1}{20.9}(O_2 + O_2') \quad (3)$$

where O_2' is the oxygen consumed and O_2 is the excess oxygen. The oxygen consumed is obtained as follows:

$$O_2' = CO_2 + \frac{1}{2}CO + \frac{1}{2}H_2O \quad (4)$$

Substituting the value of H_2O of equation (2) in equation (4) and then the resulting value of O_2' in equation (3), we have

$$N_2 = 3.785O_2 + 3.785CO_2 + 1.892CO - 1.892H_2 - 3.785CH_4 + 11.268K(CO_2 + CO + CH_4) \quad (5)$$

Nitrogen may also be determined by difference.

$$N_2 = 100 - O_2 - CO_2 - CO - H_2 - CH_4 \quad (6)$$

From equations (5) and (6), we obtain

$$O_2 + CO_2(1 + 2.355K) + CO(0.604 + 2.355K) - 0.186H_2 - CH_4(0.582 - 2.355K) = 20.9 \quad (7)$$

Equation (7) is the general equation of the theoretical relationships among the products of combustion of hydrocarbons.

The reciprocal (η) of the excess-air coefficient is obtained as follows:

$$\eta = \frac{O_2 \text{ required for complete combustion}}{O_2 \text{ present}}$$

$$\text{then } \eta = \frac{CO_2 + CO + \frac{1}{2}H_2 + \frac{1}{2}H_2O + 2CH_4}{O_2 + CO_2 + \frac{1}{2}CO + \frac{1}{2}H_2O} \quad (8)$$

Substituting the value of H_2O of equation (2), we have

$$\eta = \frac{(1 + 2.977K)(CO_2 + CO + CH_4) + [O_2 + CO_2(1 + 2.977K) + CO(0.5 + 2.977K) - CH_4(1 - 2.977K) - \frac{1}{2}H_2]}{O_2 + CO_2 + \frac{1}{2}CO + \frac{1}{2}H_2O} \quad (9)$$

Equation (9) is the general equation of the theoretical relationships of the excess-air coefficient and the products of combustion of hydrocarbons.

Inspection of table II shows that the H-C ratios of standard and fighting grade aviation gasolines, Auto Diesel fuel, and Laboratory Diesel fuel from exhaust-gas analyses are approximately the same and, therefore, for the development of an Ostwald combustion diagram they have been considered to have an average H-C ratio of 0.175. Likewise, the H and CH_4 relations to CO of the fuels (fig. 3) are approximately the same. Inserting these values in equation (7),

$$O_2 + 1.412CO_2 + 0.921CO = 20.937 \quad (10)$$

and equation (9) becomes

$$\eta = \frac{1.521CO_2 + 1.521CO + 0.333}{O_2 + 1.521CO_2 + 0.765CO - 0.105} \quad (11)$$

The combustion triangle is formed from the solution of equation (10) for CO_2 , CO , and O_2 and is plotted in figure 6.

By means of equations (10) and (11), equations for CO_2 , CO , and O_2 in terms of η may be obtained. The solution of these equations for a common value of η makes it possible to construct the η lines in the Ostwald diagram (fig. 6).

From the H-C ratio and the stoichiometric equations for complete combustion, the quantity of air required was calculated to be 14.89 pounds per pound of fuel and the air-fuel ratio was $14.89/\eta$.

REFERENCES

1. Fenning, Robert W.: The Composition of the Exhaust from Liquid-Fuel Engines. Jour. Inst. Mech. Eng., London, March 1916, pp. 185-236.
2. Judge, Arthur W.: The Testing of High Speed Internal Combustion Engines. D. Van Nostrand Co., 1925.

3. Shepherd, Martin: An Improved Apparatus and Method for the Analysis of Gas Mixtures by Combustion and Absorption. *Bur. Standards Jour. Research*, vol. 6, no. 1, January 1931, pp. 121-167.
4. Hourwich, Iskander, and Foster, W. J.: *Air Service Engine Handbook*. 1925.
5. Dicksee, C. B.: Exhaust Gas Analysis. *Auto. Eng.*, vol. 21, no. 285, October 1931, pp. 401-406.
6. Best, H. W.: Report on Air-Fuel-Ratio Tests. *S.A.E. Jour.*, vol. 25, no. 5, November 1929, pp. 532-534.
7. Theodorsen, Theodore: I: On the Propagation of Large Disturbances in a Gas. II: On the Combustion of Oil. Johns Hopkins University, Baltimore, 1929.

BIBLIOGRAPHY

- Allcut, E. A.: Further Tests on a Two-Stroke Cycle Oil Engine. *Proc. Inst. Mech. Eng.*, vol. 1, 1927, pp. 519-536.
- Burrell, George A., and Seibert, Frank M. Sampling and Examination of Mine Gases and Natural Gas. (Revision of Bull. 42) by G. W. Jones. U.S. Dept. of Commerce, Bur. Mines, Bull. 197, 1926.
- Camp, J. M.: Methods of the Chemists of the United States Steel Corporation for the Sampling and Analysis of Gases. Carnegie Steel Co., Pittsburgh, Pa., 1927.
- Clerk, D., and Burls, G. A.: The Gas, Petrol and Oil Engine. Vol. 2, 1913, John Wiley and Sons, Inc., pp. 620-635.
- Dennis, L. M., and Nichols, M. L.: Gas Analysis. The Macmillan Co., 1929.

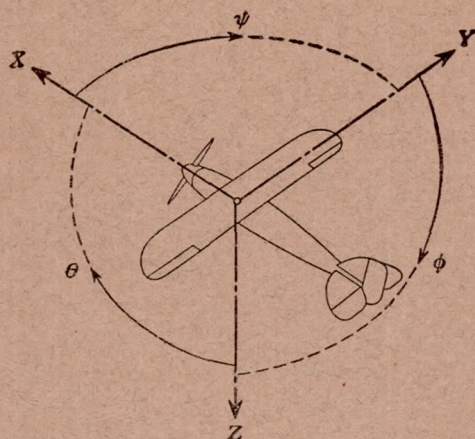
- Fieldner, A. C., Straub, A. A., and Jones, G. W.: Gasoline Losses Due to Incomplete Combustion in Motor Vehicles. *Jour. Indus. Eng. Chem.*, vol. 13, no. 1, January 1921, pp. 51-58.
- Graf, S. H., and Gleeson, G. W.: Adjustment of Automotive Carburetors for Economy. *Eng. Exper. Sta.*, Oregon State Agricultural College, Circular Series, no. 2, January 1930.
- Howe, J. M. L., and Lewis, G. W.: Study of Fuel Mixture of a Franklin Air Cooled Motor. Sibley College, Cornell University. 1908.
- Kobe, A. Kenneth: Analysis of Three Hydrocarbons by Combustion. *Indus. Eng. Chem.*, vol. 3, no. 3, Analytical Edition, July 15, 1931, pp. 262-264.
- Lockwood, E. H.: Exhaust-Gas-Analysis Calculations. *S.A.E. Trans.*, pt. 2, 1927, pp. 21 and 22.
- Lockwood, E. H.: Exhaust-Gas-Analysis Calculations. *S.A.E. Trans.*, vol. 23, 1928, pp. 373 and 374.
- Minter, Clarke C.: Interpretation of Exhaust Gas. *S.A.E. Jour.*, vol. 22, no. 1, January 1928, pp. 19-23.
- Nichols, J. T.: Saving Gasoline by Accurate Carburetor Adjustment. *Instruments*, vol. 2, no. 8, August 1929, pp. 265-267.
- Rabazzana, Hector, and Kalmar, Stephen: Investigating Mixture Distribution. *Auto. Eng.*, vol. 22, no. 301, December 1932, pp. 587-593.
- Theodorsen, Th.: Analytiske betragtninger over rökgas-analyse, samt anmerkninger til bruken av Ostwaldiagrammet. (Unpublished paper.) Norwegian Institute of Technology, Trondhjem, Norway, 1923.
- White, Alfred H.: Gas and Fuel Analysis. McGraw-Hill Book Co., 1920.

TABLE I
ENGINE-TEST CONDITIONS

Engine no.	1	2	3	4	5
Number of cylinders.....	1.....	1.....	1.....	1.....	9.....
Displacement, cubic inches.....	137.....	137.....	137.....	143.....	1,750.....
Combustion chamber.....	Pent roof.....	Bulb prechamber.....	Vertical disk.....	Pent roof.....	Spherical.....
Ignition.....	Spark.....	Compression.....	Compression.....	Spark.....	Spark.....
Cooling.....	Liquid.....	Liquid.....	Liquid.....	Liquid.....	Air.....
Fuel system.....	Carburetor and injection.....	Injection.....	Injection.....	Carburetor and injection.....	Carburetor.....
Fuel.....	Standard grade aviation gasoline.....	Auto diesel and laboratory diesel fuel.....	Auto diesel fuel.....	Standard grade aviation gasoline and safety fuel.....	Fighting grade aviation gasoline.....
Compression ratio.....	5.5.....	13.5.....	13.4-15.3.....	5.8-7.0.....	5.1.....
Engine speed, r.p.m.....	1,500.....	1,500.....	1,500.....	1,500-2,100.....	1,475-1,550.....

TABLE II
COMPOSITION OF FUELS BY WEIGHT

	Auto diesel fuel		Laboratory diesel fuel		Aviation gasoline			Hydrogenated safety fuel	
	Manufacturer's report	Average, 17 laboratory gas analyses	Manufacturer's report	Average, 10 laboratory gas analyses	Reference 4, p. 461	Fighting grade Average, 13 laboratory gas analyses	Standard grade Average, 20 laboratory gas analyses	Manufacturer's report	Average, 43 laboratory gas analyses
C.....		85.3		85.5	85.1	85.1	84.5	88.49	88.65
H.....		14.7		14.5	14.9	14.9	15.5	11.51	11.35
H/C.....		.172		.169	.175	.175	.183	.130	.128



Positive directions of axes and angles (forces and moments) are shown by arrows

Axis			Moment about axis			Angle		Velocities	
Designation	Sym- bol	Force (parallel to axis) symbol	Designation	Sym- bol	Positive direction	Designa- tion	Sym- bol	Linear (compo- nent along axis)	Angular
Longitudinal---	X	X	rolling-----	L	Y → Z	roll-----	φ	u	p
Lateral-----	Y	Y	pitching-----	M	Z → X	pitch-----	θ	v	q
Normal-----	Z	Z	yawing-----	N	X → Y	yaw-----	ψ	w	r

Absolute coefficients of moment

$$C_l = \frac{L}{qbS}$$

$$C_m = \frac{M}{qcS}$$

$$C_n = \frac{N}{qbS}$$

Angle of set of control surface (relative to neu-
tral position), δ . (Indicate surface by proper
subscript.)

4. PROPELLER SYMBOLS

D , Diameter.

p , Geometric pitch.

p/D , Pitch ratio.

V' , Inflow velocity.

V_s , Slipstream velocity.

T , Thrust, absolute coefficient $C_T = \frac{T}{\rho n^2 D^4}$

Q , Torque, absolute coefficient $C_Q = \frac{Q}{\rho n^2 D^5}$

P , Power, absolute coefficient $C_P = \frac{P}{\rho n^3 D^5}$.

C_s , Speed power coefficient $= \sqrt[5]{\frac{\rho V^5}{P n^2}}$.

η , Efficiency.

n , Revolutions per second, r. p. s.

Φ , Effective helix angle $= \tan^{-1} \left(\frac{V}{2\pi r n} \right)$

5. NUMERICAL RELATIONS

1 hp. = 76.04 kg/m/s = 550 lb./ft./sec.

1 kg/m/s = 0.01315 hp.

1 mi./hr. = 0.44704 m/s

1 m/s = 2.23693 mi./hr.

1 lb. = 0.4535924277 kg.

1 kg = 2.2046224 lb.

1 mi. = 1609.35 m = 5280 ft.

1 m = 3.2808333 ft.

